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(54) **HIGH TEMPERATURE SPLIT-FACE PROBE**

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(51) **Int. Cl.**

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F01D 11/20	(2006.01)
F01D 17/06	(2006.01)
F01D 21/00	(2006.01)

(52) **U.S. Cl.**

CPC ***F01D 11/20*** (2013.01); ***F01D 17/06*** (2013.01); ***F01D 21/003*** (2013.01)

(58) **Field of Classification Search**

CPC ... F01D 17/02; F01D 21/003; F05D 2260/80; F05D 2270/114

USPC 73/112.01

See application file for complete search history.

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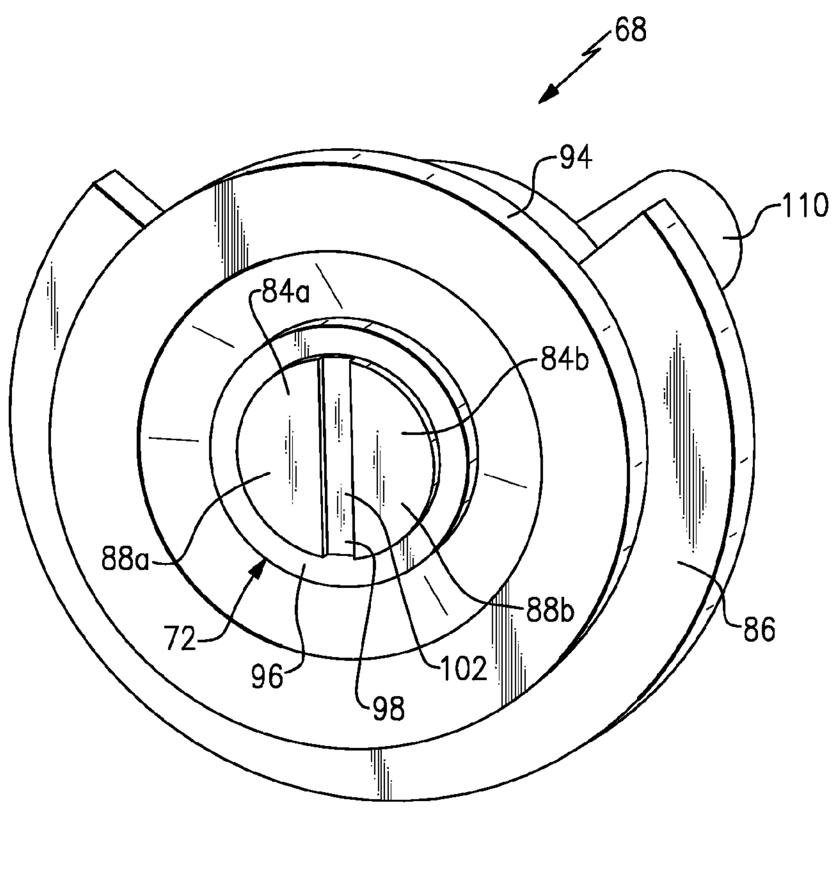
(57) **ABSTRACT**

An example split-face probe includes a sensor component having a split-face, a housing arranged about the sensor component, and at least one ceramic fitting that supports the sensor component.

18 Claims, 4 Drawing Sheets



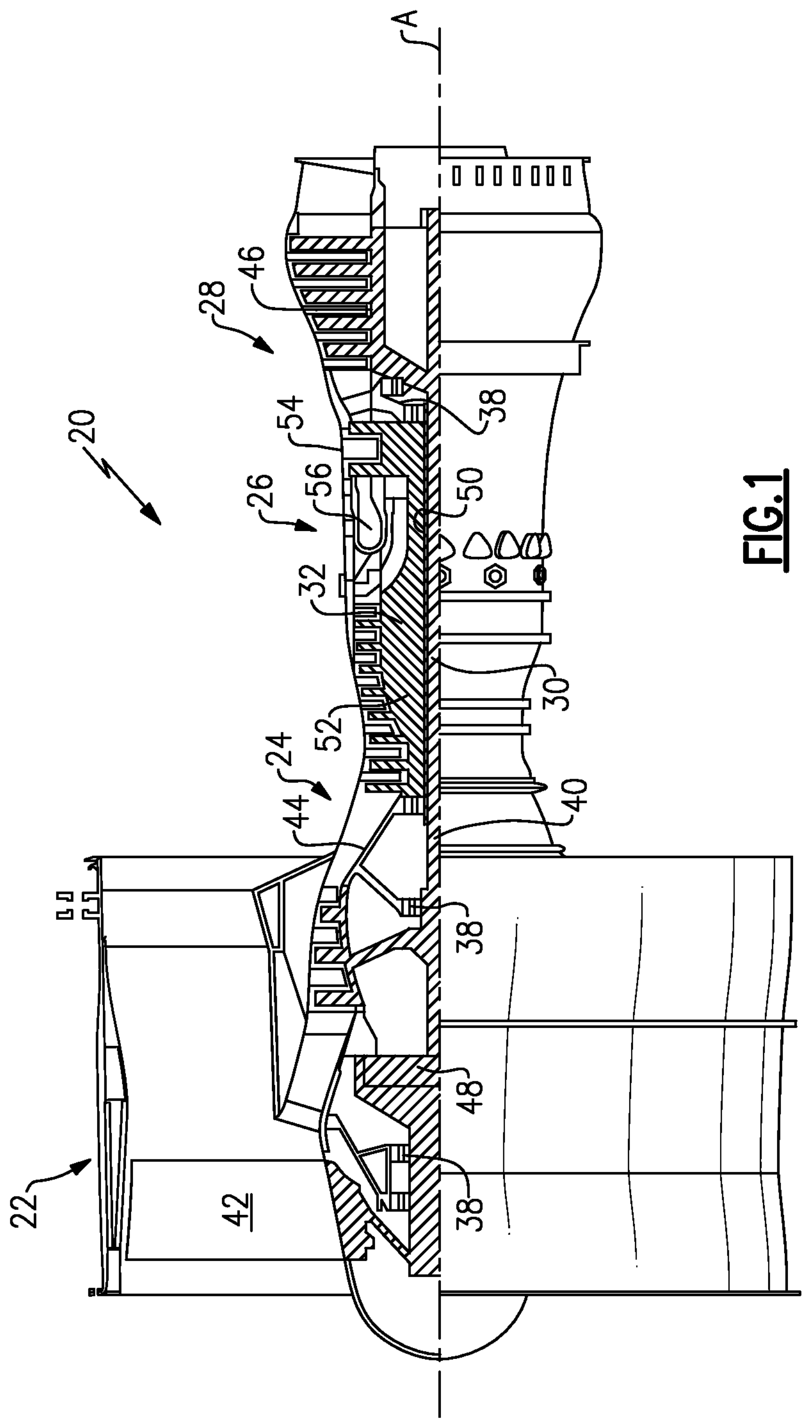


FIG. 1

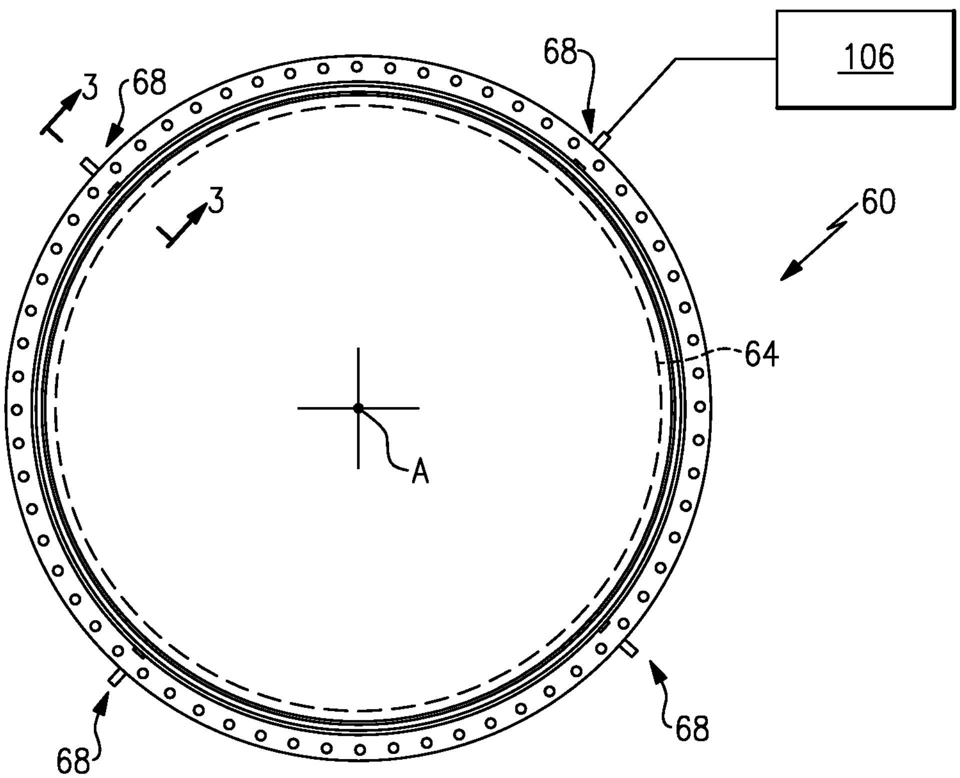


FIG.2

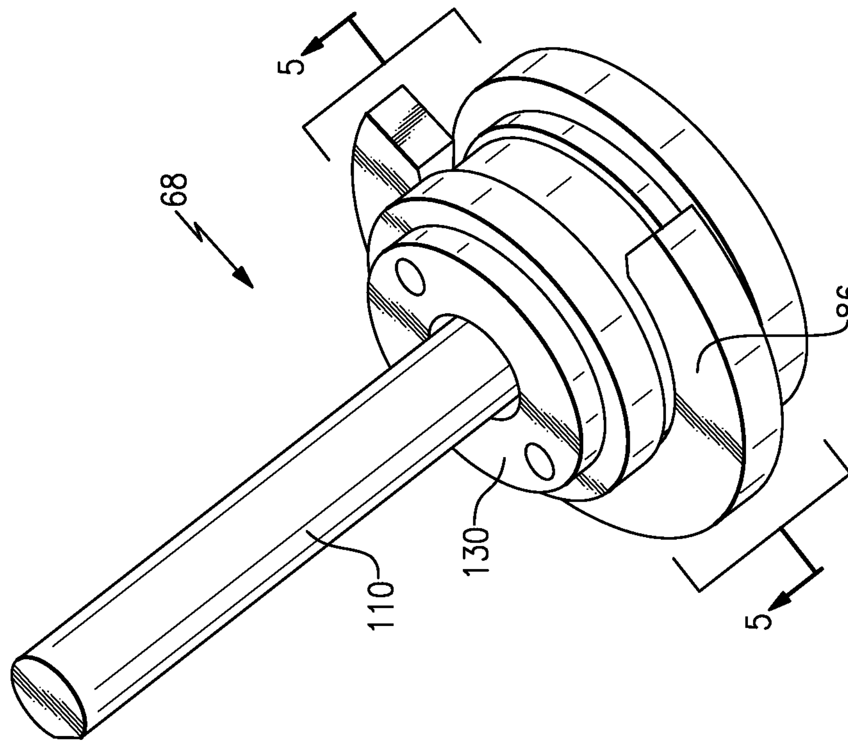


FIG.4

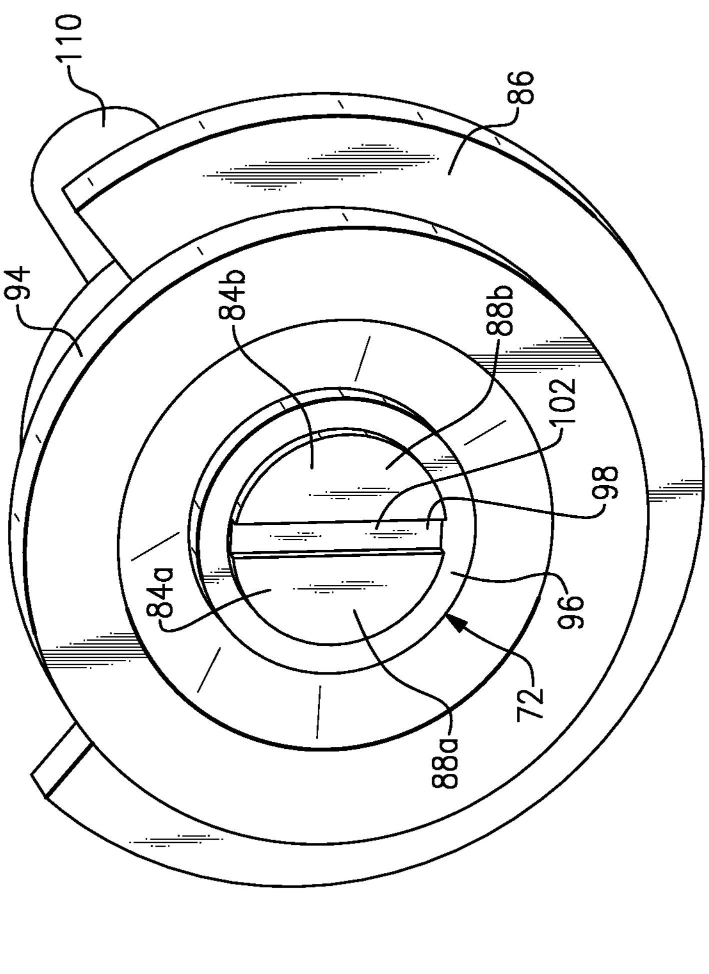


FIG.3

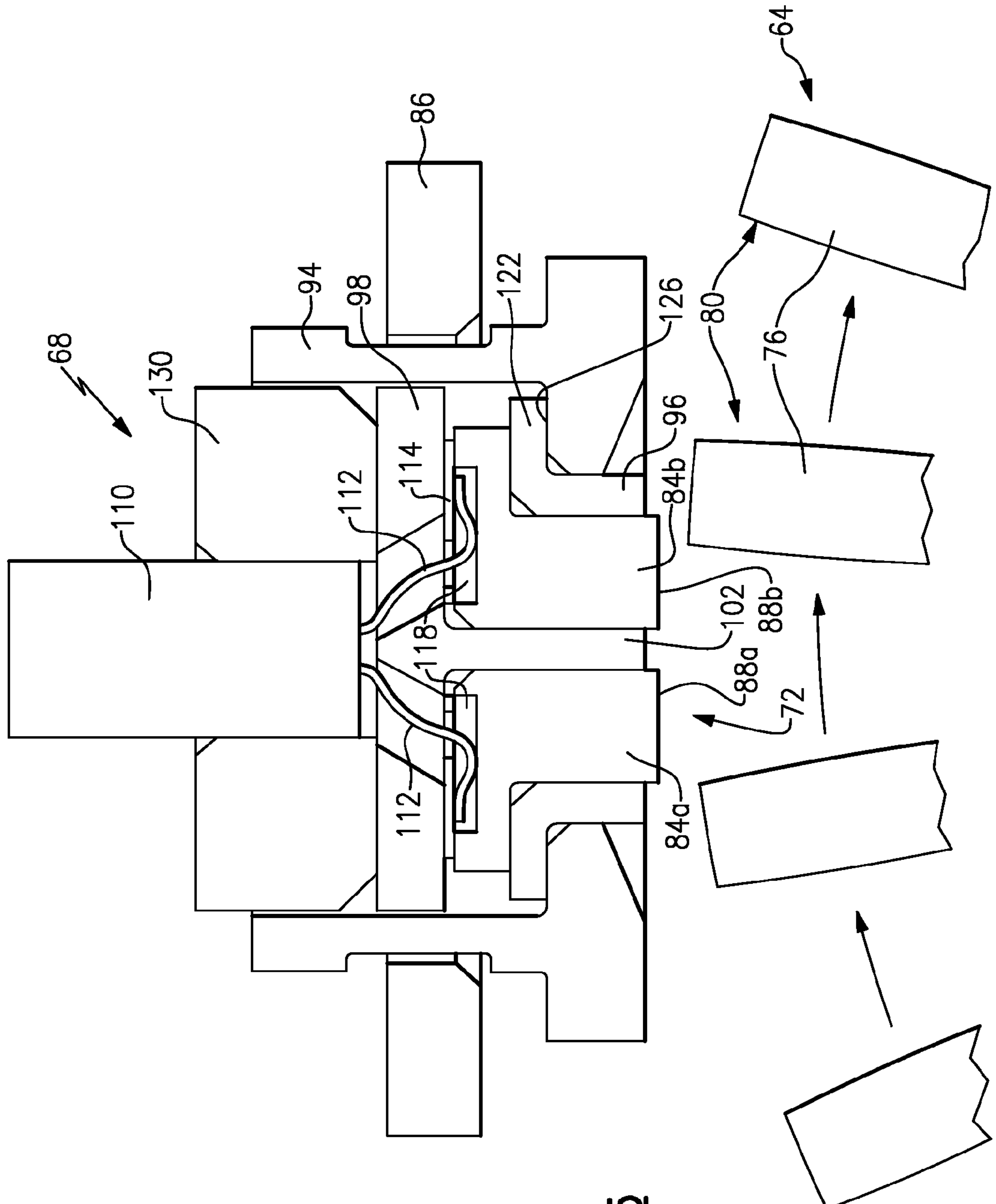

68
130
110
112
114
98
94
86
122
126
96
84b
80
76
64
102
88b
88a
72
118
112
84a

FIG.5

HIGH TEMPERATURE SPLIT-FACE PROBE

BACKGROUND

This disclosure relates generally to a measurement probe and, more particularly, to a split-face capacitance probe used in high temperature environments, such as environments having temperatures above 400° F.

Turbomachines, such as gas turbine engines, typically include a fan section, a compression section, a combustion section, and a turbine section. Turbomachines may employ a geared architecture connecting portions of the compression section to the fan section.

The turbomachine may include an annular case structure that circumscribes a rotatable array of blades. Tip-timing probes mounted to the case can be used to monitor vibratory stresses within the blades. Tip clearance probes detect tip clearance to the blades. Split-face capacitance probes with circuit boards have been used in areas of the turbomachine having a relatively low temperatures, such as temperatures at or below 400° F. (204° C.). These probes may become damaged if used in other, higher temperature environments of the engine.

SUMMARY

A split-face probe according to an exemplary aspect of the present disclosure includes, among other things, a sensor component having a split-face. A housing is arranged about the sensor component. At least one ceramic fitting supports the sensor component.

In a further nonlimiting embodiment of the foregoing split-face probe, the sensor is configured for use at temperatures above 400° F.

In a further nonlimiting embodiment of either of the foregoing split-face probes, the sensor component may include individual sensors separated from each other by a portion of the at least one ceramic fitting.

In a further nonlimiting embodiment of any of the foregoing split-face probes, the portion of the at least one ceramic fitting bisects the split-face.

In a further nonlimiting embodiment of any of the foregoing split-face probes, the at least one ceramic fitting includes an upper ceramic, a lower ceramic, and a portion of the sensor component sandwiched therebetween.

In a further nonlimiting embodiment of any of the foregoing split-face probes, at least one strap electrically couples the sensor component to a hard lead. The at least one strap is sandwiched between the sensor component and the upper ceramic.

In a further nonlimiting embodiment of any of the foregoing split-face probes, the sensor component is supported exclusively by the at least one ceramic fitting.

In a further nonlimiting embodiment of any of the foregoing split-face probes, the sensor component is a capacitance-based sensor component.

In a further nonlimiting embodiment of any of the foregoing split-face probes, the at least one ceramic fitting circumscribes the split-face.

In a further nonlimiting embodiment of any of the foregoing split-face probes, the at least one ceramic fitting comprises aluminum oxide.

A method of detecting a blade-related measurement includes supporting a split-face sensor component with at least one ceramic fitting.

In a further nonlimiting embodiment of the foregoing method of detecting, the split-face sensor component is configured for operation in environments having temperatures exceeding 400° F.

In a further nonlimiting embodiment of either of the foregoing methods of detecting, the split-face sensor component comprises a capacitance sensor.

In a further nonlimiting embodiment of any of the foregoing methods of detecting, the split-face sensor component is configured to detect the time of arrival of a turbomachine blade tip.

A turbomachine according to another exemplary aspect of the present disclosure includes, among other things, a gas path having a plurality of rotors and stators. A probe is configured to detect a turbomachine blade-related measurement. The probe includes a sensor component having a split-face, a housing arranged about the sensor component, and at least one ceramic fitting that supports the sensor component.

In a further nonlimiting embodiment of the foregoing turbomachine, the sensor component is configured for use at temperatures above 400° F.

In a further nonlimiting embodiment of either of the foregoing turbomachines, the sensor component includes individual sensors separated from each other by portion of the at least one ceramic fitting.

In a further nonlimiting embodiment of any of the foregoing turbomachines, the portion of the at least one ceramic fitting bisects the split-face.

In a further nonlimiting embodiment of any of the foregoing turbomachines, the sensor component is a capacitance-based sensor component.

In a further nonlimiting embodiment of any of the foregoing turbomachines, the at least one ceramic fitting circumscribes the split-face.

DESCRIPTION OF THE FIGURES

The various features and advantages of the disclosed examples will become apparent to those skilled in the art from the detailed description. The figures that accompany the detailed description can be briefly described as follows:

FIG. **1** shows an example turbomachine.

FIG. **2** shows an aft view of a compressor case of the turbomachine of FIG. **1**.

FIG. **3** shows perspective view of a split-face probe held within the case of FIG. **2**.

FIG. **4** shows another perspective view of the split-face probe of FIG. **3**.

FIG. **5** shows a section view at line **5-5** in FIG. **4**.

DETAILED DESCRIPTION

FIG. **1** schematically illustrates an example turbomachine, which is a gas turbine engine **20** in this example. The gas turbine engine **20** is a two-spool turbofan gas turbine engine that generally includes a fan section **22**, a compression section **24**, a combustion section **26**, and a turbine section **28**.

Although depicted as a two-spool turbofan gas turbine engine in the disclosed non-limiting embodiment, it should be understood that the concepts described herein are not limited to use with turbofans. That is, the teachings may be applied to other types of turbomachines and turbine engines including three-spool architectures. Further, the concepts described herein could be used in environments other than a turbomachine environment and in applications other than aerospace applications.

In the example engine **20**, flow moves from the fan section **22** to a bypass flowpath. Flow from the bypass flowpath generates forward thrust. The compression section **24** drives air along the core flowpath. Compressed air from the compression section **24** communicates through the combustion section **26**. The products of combustion expand through the turbine section **28**.

The example engine **20** generally includes a low-speed spool **30** and a high-speed spool **32** mounted for rotation about an engine central axis A. The low-speed spool **30** and the high-speed spool **32** are rotatably supported by several bearing systems **38**. It should be understood that various bearing systems **38** at various locations may alternatively, or additionally, be provided.

The low-speed spool **30** generally includes a shaft **40** that interconnects a fan **42**, a low-pressure compressor **44**, and a low-pressure turbine **46**. The shaft **40** is connected to the fan **42** through a geared architecture **48** to drive the fan **42** at a lower speed than the low-speed spool **30**.

The high-speed spool **32** includes a shaft **50** that interconnects a high-pressure compressor **52** and high-pressure turbine **54**.

The shaft **40** and the shaft **50** are concentric and rotate via bearing systems **38** about the engine central longitudinal axis A, which is collinear with the longitudinal axes of the shaft **40** and the shaft **50**.

The combustion section **26** includes a circumferentially distributed array of combustors **56** generally arranged axially between the high-pressure compressor **52** and the high-pressure turbine **54**.

In some non-limiting examples, the engine **20** is a high-bypass geared aircraft engine. In a further example, the engine **20** bypass ratio is greater than about six (6 to 1).

The geared architecture **48** of the example engine **20** includes an epicyclic gear train, such as a planetary gear system or other gear system. The example epicyclic gear train has a gear reduction ratio of greater than about 2.3 (2.3 to 1).

The low-pressure turbine **46** pressure ratio is pressure measured prior to inlet of low-pressure turbine **46** as related to the pressure at the outlet of the low-pressure turbine **46** prior to an exhaust nozzle of the engine **20**. In one non-limiting embodiment, the bypass ratio of the engine **20** is greater than about ten (10 to 1), the fan diameter is significantly larger than that of the low pressure compressor **44**, and the low-pressure turbine **46** has a pressure ratio that is greater than about 5 (5 to 1). The geared architecture **48** of this embodiment is an epicyclic gear train with a gear reduction ratio of greater than about 2.5 (2.5 to 1). It should be understood, however, that the above parameters are only exemplary of one embodiment of a geared architecture engine and that the present disclosure is applicable to other gas turbine engines including direct drive turbofans.

In this embodiment of the example engine **20**, a significant amount of thrust is provided by the bypass flow B due to the high bypass ratio. The fan section **22** of the engine **20** is designed for a particular flight condition—typically cruise at about 0.8 Mach and about 35,000 feet. This flight condition, with the engine **20** at its best fuel consumption, is also known as “Bucket Cruise” Thrust Specific Fuel Consumption (TSFC). TSFC is an industry standard parameter of fuel consumption per unit of thrust.

Fan Pressure Ratio is the pressure ratio across a blade of the fan section **22** without the use of a Fan Exit Guide Vane system. The low Fan Pressure Ratio according to one non-limiting embodiment of the example engine **20** is less than 1.45 (1.45 to 1).

Low Corrected Fan Tip Speed is the actual fan tip speed divided by an industry standard temperature correction of Temperature divided by 518.7^0.5. The Temperature represents the ambient temperature in degrees Rankine. The Low Corrected Fan Tip Speed according to one non-limiting embodiment of the example engine **20** is less than about 1150 fps (351 m/s).

Referring now to FIGS. **2** to **5** with continuing reference to FIG. **1**, an example case is a compressor case **60** from the high-pressure compressor section **52** of the engine **20**. The compressor case **60** circumscribes a compressor blade array **64**. For clarity, the compressor blade array **64** is shown in broken line form in FIG. **2**.

The compressor case **60** includes several split-face probes **68** that include sensor components **72**. The example probes **68** are measurement probes used to measure blade-related measurements. For example, the probes **68** may measure the time of arrival and thereby deflection and stress of blades **76** of the array **64**. Specifically, the probes **68** may measure a circumferential position of tips **80** of the blades **76** as the array **64** is rotated relative to the probes **68** during operation of the engine **20**. The actual circumferential position of the tips **80** is compared to a predicted position of the tips **80** to determine deflection of the blades **76**, which may help indicate stress on the blades **76**.

The example sensor component **72** of the probe **68** is a metallic capacitance-based sensor. The sensor component **72** includes a first sensor **84*a*** having a hemispherical sensor face **88*a***, and a second sensor **84*b*** having a hemispherical sensor face **88*b***. A housing **94** is arranged about the first and second sensors **84*a*** and **84*b***. A retaining member **86** member **90** may be used to hold the probe **68** to the case compressor **60**.

The probes **68** may be dual-measurement probes that also measure radial clearance between the tips **80** of the blades **76** and the sensor faces **88*a*** and **88*b***. Clearance is the radial distance between the tips **80** and the faces **88*a*** and **88*b*** and can be detected by changes in amplitude of a signal from the sensor component **72**.

The sensors **84*a*** and **84*b*** are reversed in polarity and sandwiched radially between a first ceramic fitting **96** and a second ceramic fitting **98** within the housing **94**. The first example ceramic fitting **96** is a lower ceramic that circumscribes a portion of the sensor component **72**.

The second ceramic fitting **98** is an upper ceramic in this example. The second ceramic fitting **98** includes a flange **102** that extends radially between the first sensor **84*a*** and a second sensor **84*b***. The flange **102** bisects the hemispherical sensor faces **88*a*** and **88*b***. The flange **102** provides a zero-crossing voltage signal. Tip-timing is typically the circumferential time-of-arrival and can be extracted from the time of the zero-crossing of the signal.

The first sensor **84*a*** and **84*b*** are operably coupled to a controller **106** through a hard lead **110**, conductor wires **112**, and straps **114**. These components help connect the split-face probe **68** to a capacitance to voltage converter circuit. The controller **106** may include a signal conditioner.

The first and second sensors **84*a*** and **84*b*** include a groove **118** that accommodate portions of the conductor wires **112**. Sandwiching the sensors **84*a*** and **84*b*** between the first ceramic fitting **96** and the second ceramic fitting **98** urges the conductor wires **112** against the straps **114**, which operably connects the first and second sensors **84*a*** and **84*b*** to the hard lead **110**.

The first ceramic fitting **96** includes an annular flange **122** that rests against a shoulder **126** of the housing **94** to limit radially inward movement of the first ceramic fitting **96**. A cap

130 may be welded, press- or interference-fit into the housing **94** to limit radially outward movement of the second ceramic fitting **98**.

In this example, exclusively the first ceramic fitting **96** and the second ceramic fitting **98** support the sensors **84***a* and **84***b*. The first ceramic fitting **96** and the second ceramic fitting **98** also electrically isolate and insulate the sensors **84***a* and **84***b*, from the housing **94**, the cap **130**, the retaining member **86** (which all may be steel).

In this example, the first ceramic fitting **96** and the second ceramic fitting **98** are both aluminum oxide (or alumina) material, such as a 99.5 percent pure Al_2O_3.

Features of the disclosed examples include a split-face probe that is suitable for use at temperatures above 400° F. and as high as 1400° F. The split-face probes in the prior art includes circuit board material, which can become damaged at such temperatures.

The preceding description is exemplary rather than limiting in nature. Variations and modifications to the disclosed examples may become apparent to those skilled in the art that do not necessarily depart from the essence of this disclosure. Thus, the scope of legal protection given to this disclosure can only be determined by studying the following claims.

I claim:

1. A split-face probe comprising:

a sensor component having a split-face;

a housing arranged about the sensor component; and

at least one ceramic fitting that supports the sensor component and circumscribes the split-face, wherein the at least one ceramic fitting is electrically non-conductive.

2. The probe of claim **1**, wherein the sensor is configured for use at temperatures above 400 degrees Fahrenheit.

3. The probe of claim **1**, wherein the sensor component includes individual sensors having sensing faces that are configured to face outwardly from the split-face probe toward a blade array, the sensing faces completely separated from each other by a portion of the at least one ceramic fitting.

4. The probe of claim **3**, wherein the portion of the at least one ceramic fitting bisects the split-face.

5. The probe of claim **1**, wherein the at least one ceramic fitting includes an upper ceramic, a lower ceramic, and a portion of the sensor component sandwiched therebetween.

6. The probe of claim **5**, including at least one strap that electrically couples each sensor component to a hard lead, at least one strap sandwiched between the sensor component and the upper ceramic.

7. The probe of claim **1**, wherein the sensor component is supported exclusively by the at least one ceramic fitting.

8. The probe of claim **1**, wherein the individual sensors each include a sensing face, wherein the sensing face of each of the individual sensors is separated from the sensing face of the other individual sensors by the at least one ceramic fitting.

9. The probe of claim **1**, wherein the split-face is separated by the portion of the at least one ceramic fitting into two separate and distinct semi-circular sensing faces that are completely separated from each other by the portion of the at least one ceramic fitting.

10. A method for detecting a blade related measurement comprising:

Supporting a split-face sensor component with at least one ceramic fitting,

Said fitting circumscribes the split-face, wherein the at least one ceramic fitting is electrically non-conductive.

11. The method of claim **10**, wherein the split-face sensor component is configured for operation in environments having temperatures exceeding 400 degrees Fahrenheit.

12. The method of claim **10**, wherein the split-face sensor component is configured to detect a turbomachine blade tip.

13. The method of claim **10**, wherein the split-face includes at least two sensing faces spaced from each other by a portion of the at least one ceramic fitting.

14. A turbomachine comprising:

a gas path including a plurality of rotors and stators; and

a probe configured to detect a turbomachine blade-related measurement, the probe comprising,

a sensor component having a split-face,

a housing arranged about the sensor component,

at least one ceramic fitting that supports the sensor component and circumscribes the split-face, wherein the at least one ceramic fitting is electrically non-conductive.

15. The turbomachine of claim **14**, wherein the sensor component is configured for use at temperatures above 400 degrees Fahrenheit.

16. The turbomachine of claim **14**, wherein the sensor component includes individual sensors that each have one of a plurality of forward facing sensing faces, the plurality of forward facing sensor faces are each completely separated from each other by a portion of the at least one ceramic fitting.

17. The turbomachine of claim **14**, wherein the portion of the at least one ceramic fitting bisects the split-face.

18. The turbomachine of claim **14**, wherein the split-face comprises two front sensing faces completely separated and spaced from each other by a portion of the at least one ceramic fitting.

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